

# AVIATION

AUGUST 20, 1923

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Navy-Wright seaplane entered in Schneider Cup race which made 177.5 m.p.h. in recent trials

VOLUME  
XV

## SPECIAL FEATURES

THE "TURKEY BUZZARD" GLIDER  
FRENCH FUEL CONSUMPTION AIR RACES  
FIGHTING BOLL WEEVIL WITH AIRPLANES  
PRINCIPLES OF EFFICIENT AILERON DESIGN

NUMBER  
8

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# WRIGHT

AUGUST 30, 1923

# AVIATION

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No. 8

### The Start of Night Flying

THIS month will probably mark the beginning of a new era in aviation. The night flying service of the Air Mail is scheduled to begin operations in the latter part of the month and the eyes of the entire aeronautical world are turned to their attention in Omaha.

Over 100,000,000 years ago, when the Wright brothers made their first public flights, there has been no rest, in our opinion, that has resulted such an advance in the practical application of the airplane in commercial purposes as the introduction of the twenty-four-hour-a-day flying.

It is not too much to say that the United States Air Mail service is the only organization in the world that could undertake an experiment with good chances of success. Since the Air Mail Service was started by Otto Prager, then Special Assistant Postmaster General, its record has been a series of successes prior to this country. He looked at mail carrying in relation from the Post Office viewpoint. He knew that to be worth while the Air Mail must achieve reliability and regularity. Delivered in the winter, thunderstorms in the summer, fog, snow and wind have all been overcome by the devoted pilots of the Air Mail, supported by an inspection and maintenance service which is probably the best in the world. Mr. Prager will always receive the credit and honor due to pioneers in any hazardous enterprise.

For that reason, he has picked up the thread of advancement and through his far-sighted enthusiasm has organized and developed the night flying idea into what will soon be a reality. Men of this mold are of the staff that makes the results lead to no many fields of endeavor.

When night flying was first considered, it was not left to the test of a pilot or to the reliability of an engine to make a test of the scheme. Instead, it was approached from an engineering angle. The route was surveyed by experts, engineering groups were invited from the Air Mail pilots and others experienced in cross country flying. Finally, the lighting systems of the country were called on for their best equipment. Orders were placed with these manufacturers for new airplanes specially designed for this work, and the designers of these flights were not hampered by petty restrictions. They were told what the Air Mail Service wanted the airplanes to do and were left to work out the problem for themselves.

When night flying is begun, it will be an unqualified success. It is not necessary to make it at the present time, for it was formerly planned in advance, which reduces the fortunate element to a minimum. There are bound to be the difficulties, but are already mentioned in any new enterprise. However, these will be overshadowed by the daily proof, for nothing should be allowed to weaken the spirit of the employees Air Mail people. Congress, too, should uphold

the hands of these men of vision and common sense whose endeavor will bring to the United States another achievement of world wide import.

### Commercial Air Transport

THE high hopes for commercial air transport that were entertained for a year or so after the war have not materialized. For a while, there was hardly a month that did not bring the news of another enterprise for the development of some air transport service. The stock market moved on commercial aviation as a bubble, and it is to be noted the losses sustained by reckless and unscrupulous investors.

The Aeroexpress Airways has made a token effort to pioneer the water savings, and General Loomis's Northwest enterprise was a creditable beginning. But outside of these two air lines little, except glowing newspaper publicity, has come from the promotion of air lines.

A subsidy, private or governmental, now appears to be necessary to do for commercial air transport what was done for the railroads in the preceding days. Both of these terms of support are difficult to secure, and the outlook is freshly discouraging.

Some progress has been made in electricity, the Ford in radio, and Edison in the steamship, may perhaps come forth and make some startling discovery or improvement that will do for the airplane what those men did in their chosen lines. Until then, a hopeful spirit must be abandoned by those in aviation, confidently looking to the future from the present envelope puff of activity.

### Fuel Consumption Races

AN innovation at the racing enterprises has been made in France with the organization of airplane races based on fuel consumption. As our readers may read in this issue in one of these events, the French Cup, each widely different shape at a light plane and a multi-engine plane were pitted against one another. In this, as in another race, open to light planes, fuel economy was the basic factor determining the placing of the contestants.

This enterprise is worthy to be commended, for the cost of commercial aviation largely depends on the airplane's fuel load and this again is governed to a very great extent by the fuel consumption of the engine. Inasmuch as for every pound of gasoline and oil used, a pound of payload must be carried, the importance of this question becomes apparent. It is hoped that contests embodying this feature may be expected at an early date in this country.





ward and thickness toward the tips. No horizontal stabilizer is fitted.

The Clerget engine fitted to one of these ships differs in several particulars from the power plant used in the English and French versions, which is fitted to the other two. The new engine has a bore of 50 mm., a stroke of 120 mm. and develops a maximum of 15 hp. at 1800 r.p.m., although its normal rating is 1000 r.p.m., giving about 10 hp. The cylinder bearings have been substituted for the ball bearings originally used, and a pressure lubrication system, with an oil pump, installed. The intake manifold leads through the lower part of the cowling which acts as an oil sump, so that the fuel is heated before admission.

The Delorme engine used on one of the Dewandins is the same as that fitted to the Farman. The Voisin engine, with which the third Dewandins ship was equipped, is a four-cylinder, under horizontal opposed type, cooled by air, which develops 12 hp. at 1400 r.p.m. and weighs complete 41.5 kg. The bore is 54 mm. and the stroke is 110 mm. The valves are operated by pushrods, and the lubrication is of the overhead type. Sixty cylinders and cast iron cylinder heads are used.

The average distribution of weights in the Dewandins light plane is as follows: glider, 110 kg.; engine, propeller, and landing gear, 40 kg.; fuel and oil for 5 hr., 20 kg.; pilot and accessories, 30 kg.; total 204 kg. The maximum speed of the ship is 100 km. per hr. and the landing speed 20 km. per hr.

### The Braguet Entry

The Braguet entry known as the "Colibri," is a conventional monoplane having wooden wings of trapezoidal form, rounded forward and tapered to the tips. The fuselage is a tube fitted with a small Braguet engine, but in the race a 16 hp. Clerget engine was also mounted. The "Colibri" has a wing area of 35 sq. m., a span of 30.4 m., an overall length of 19 m. and a weight loaded 250 kg. It was one of the lightest planes entered in the race.

The Dewandins plane is a conventional monoplane equipped with a 50 hp. A.B.C. and is cooled by air drawn in, like a ship, which carries the pilot's cockpit, and the lack of a fuselage. Two tail booms carry the empennage which consists of a double masted and a second elevator. The span is 28 m., the overall length 3.20 m., the wing area 15 sq. m. and the gross weight 250 kg. As the power plant of this ship was not of French manufacture, and the constructor failed time to substitute a French engine, it did not participate in the race. The ship is also entered in the Voisin glider and light plane race.

The Harriot light plane is also a conventional monoplane and has the following characteristics: span 9.60 m., overall length 4.75 m., wing area 17.5 sq. m., weight loaded 200 kg. Lack of a suitable French engine had prevented this ship from taking part in the race, as the trial flights were made with a 50 hp. Anzani engine, which is of Italian origin. The Harriot-Valliant is a monoplane with a conventionally loaded wing, 10 m. span, 5 m. overall length, 10 kg. m. wing area, and 90 kg. weight empty. A 16 hp. Anzani engine was used in the trial flights, and a French engine was also substituted, but this was not available in time. The same engine prevented the Harriot-Valliant light plane from taking part in the race. This ship measures 7 m. in span, 4 m. in overall length, and has 9 kg. m. wing area. The weight empty is 90 kg.

### Harriot Light Plane

The Harriot light plane is a biplane with staggered wings, measuring 6 m. in span, 5.00 m. in overall length, and 2 m. in height. The wing area is 12 sq. m., the weight empty 90 kg., and the weight loaded 160 kg. The power plant is a 16 hp. Gnome-Rhone.

The Delorme is a tailless canard-type monoplane of 14 m. in wing area, 5.56 m. span, 5.56 m. overall length, 1.50 m. height, and 150 sq. m. total wing area. The engine is a 10 hp. Clerget. The Canard or "Serpier" is also a tailless monoplane, having two sets of surfaces for lateral and longitudinal control. One set is adjustable for a given state. Directional control is secured by vertical fins, and the machine has a span 6 m., overall length 3.60 m., wing area 30 sq. m., weight loaded 250 kg. Engine, 10 hp. Clerget.

The Lathuysens is a monoplane with variable cowling

wings and variable engine seat plane. The wings are fixed under the cowling and hinged to the latter by means of four struts. The wing area is 5.56 sq. m., span 5 m., overall length 4.76 m., height 1.45 m., weight loaded 220 kg. The engine is an 18 hp. Clerget. This ship was sent ready for the race due to lack of funds.

### A Peyret Light Plane

The Peyret is an orthodox strut braced monoplane which is made interesting because of the use of aluminum tubing in the length of the wings, just as in the Peyret landing glider, and which are used both for lateral and longitudinal control. In the latter case the movement of the ailerons is achieved by that of an elevator, but there is no first stabilizer. The wings are built up of Duralumin wing boxes and three ribs, the covering being of reinforced linen. The ribs are spaced every 45 cm. and have withstood a load of 90 kg. without failure. The wing struts are strengthened duralumin tubes. The fuselage is made of four wooden laminations covered with veneer panels. The wing area is 35 sq. m., the span 30.4 m., overall length 5.57 m., height 1.50 m., and the weight loaded 250 kg. It had made a 25 hp. Clerget engine, the ship has a maximum speed of 100 km. per hr. and a maximum speed of 44 km. per hr. The ceiling is between 3000 and 3500 m.

The Bordenly light plane, built by the Bordenly firm in the design of M. Bordenly, the famous designer of the new racing glider, is a conventional monoplane of 14 sq. m. in wing area, 8 m. span, 5.10 m. overall length, 4.70 m. height and 245 kg. weight loaded. Lack of a suitable propeller prevented this ship from taking part in the race.

No information is on hand regarding the Collet light plane. Regarding the situation track and the race the following data will be of interest. The starting time of 500 m. at 10 m. per hour was easily achieved by the Farman, the Voisin, and the Braguet. The Farman finished 500 m. in 7 m. 25 sec., the Braguet 800 m. in 27 sec. while the pilots of the three Dewandins made the following progress: 500 m. in 18 m. 25 sec. and 500 m. in 17 sec. First 500 m. in 27 sec. and Dorel 800 m. in 35 sec. and 500 m. in 20 sec.

### How the Race was Run

As was and before, the race was run in strenuous wooded surroundings and quickly alternating with fog. The first lap of 50 km. was covered by all the seven contestants in the following times: Dorel, (Dewandins) 7 m. 43 sec.; Thout (Dewandins) 7 m. 52 sec.; Barbet (Dewandins) 8 m. 36 sec.; Dewandins (Farman) 8 m. 4 m.; Fiat (Dewandins) 8 m. 52 sec.; Dorel (Farman) 9 m. 9 sec. The race was run in the Dewandins and the Braguet were clearly because of their low speed. The latter were clearly because of their low speed in the race in one of the best wing conditions.

However, on the second lap the Farman of Dorel dropped out of the running, and on the third lap Dorel finished last. Then on the fourth lap Dorel was in the fourth lap. All three pilots had their spark plugs short circuited at a thunderstorm. The two Farman came over on landing, but without much damage, the pilots being safe. The left engine of the Farman alone against the three Dewandins, a heavy handicap, considering the higher speed of these ships. Barbet attracted much attention by the splendid landing made on his Clerget powered Dewandins, making about 13 km. per hr. Every time a Dewandins, even very close to the ground at about the same speed, while Fiat, on the third Dewandins, and Comet on the remaining Farman both five forty high with very little, making not more than 60 km. per hr.

At the end of the track line Dorel and Barbet were leading with Comet third and Fiat fourth. In the 19th lap Fiat made a second landing on account of valve trouble, and the same thing happened to Dorel in the 18th lap. Then Barbet was 40 km. ahead of the other ships. At the end of the fourth lap 2 hr. 39 m. 52 sec. on against 3 hr. 4 m. 31 sec. but Comet's Farman—Barbet's victory seemed certain, when in the 28th lap he ran out of fuel and was forced to land. The

left Clerget alone in the race which he duly finished, landing at 100 km. per hr. after having covered a 31st lap, and covering the 30 km. in 17 m. 17 sec. 10 sec. at an average speed of 147 km. per hour. The race was won by the Farman of Dorel, who had covered the 30 km. distance with 26.6 l. of gasoline and 0.5 l. of oil—approximately 90 miles to the gallon. On landing 12 l. of gasoline still remained in the tank.

Comet had won the 100,000 franc cash prize of the Petit Paris, while Barbet won the sum of 20,000 francs as a consolation prize. As the atmospheric conditions had much to do with the dropping out of the contestants, much of the engine and propeller trouble experienced before and during the race was due to short time available for tuning up the power plants.

It is already mentioned that the winning engine was de-

## France Holds Novel Fuel Consumption Race

### Zeith Cup Won by Farman F90 Commercial Transport Airplane

The first annual contest for the Zeith Cup, a trophy offered by the French manufacturers of Zeith carburators, and the fuel consumption performance with the lowest fuel consumption, held July 20 and 21, last, was won by Com. Dorel on a Farman F90 class transport plane equipped with a 300 hp. Delorme engine and Lathuysens radiator. The same ship won last Dewandins the air transport contest for the Grand Prix of Paris.

The Zeith Cup race offered from the representing viewpoint was a contest between of interest in that no fuel was allowed to be consumed, gross weight, speed or wing area of the competing planes. The rules merely provided that the competitors must not consume more than 255 kg. of fuel for every 100 km. flown, or that the average speed must not be less than 70 km. per hr., and that the Farman-Lyon-Paris legs of the race were to be covered separately. Consequently the rules enabled the competitors to choose the most economical fuel consumption. Such a provision may seem odd at first sight, but in practice it worked out all right, for the fuel limitation handicapped high powered machines that might have won

the Zeith Cup in the race which he duly finished, landing at 100 km. per hr. after having covered a 31st lap, and covering the 30 km. in 17 m. 17 sec. 10 sec. at an average speed of 147 km. per hour. The race was won by the Farman of Dorel, who had covered the 30 km. distance with 26.6 l. of gasoline and 0.5 l. of oil—approximately 90 miles to the gallon. On landing 12 l. of gasoline still remained in the tank.

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signed, built and tested in two months, but several other power plants that were mounted in the planes a few days before the race, which for some of them contained the first flying trials. That under such circumstances engine failure was never played an important role in the history of the race is only natural.

Nevertheless, such as it was, the new proved one thing: namely that light planes are by no means the dangerous playthings that are sometimes represented to be. That despite the great handicap of incompletely tuned up engines, they were capable of performing satisfactorily in a new field under gusty weather conditions, indicates the new school of thought devoted to experimental flight. The race did not preclude the ideal coach airplane as yet, but it points the way to the necessary improvements from which it will spring.

The following list shows the planes entered: (The cockpit PL, and PL indicate pilot load, including pilot, and fuel load, respectively.)

1. Farman F90 sport biplane (80 hp. Anzani). Pilot, Lathuysens. PL—100 kg., PL—75 kg.
2. Farman F90 biplane (200 hp. Delorme). Pilot, Dorel. PL—500 kg., PL—230 kg.
3. Dewandins C50 sport biplane (160 hp. Le Rhône). Pilot, Vauvrière. Did not start.
4. Dewandins F90 sport biplane (160 hp. Hispano). Pilot, Bécherey. PL—200 kg., PL—100 kg.
5. Dewandins C50 sport biplane (160 hp. Le Rhône). Pilot, Vauvrière. PL—215 kg., PL—107 kg.
6. Dewandins light plane (100 hp. Voisin). Pilot, Dorel. PL—90 kg., PL—22 kg.



The Farman F90 5-passenger cabin ship (260 hp. Delorme) which won the Zeith Cup race between Paris and Lyons, France, July 21-22.

7 Spad 809 sport biplane (120 hp. Le Rhône). Pilot Robert Joyce. Did not start.

8 Bessie/4-4. Motor monoplane (750 hp. Anzani). Pilot, Bremonet. Did not start.

9 Ponce type VIII sport biplane (90 hp. Anzani). Pilot, Jacques Rouquet. PL—124 kg.; PL—68.5 kg.

The start of the race took place in splendid weather at Orly aerodrome, well known to American pilots who visit overseas, which is situated South of Paris. The start was given from 11:30 a. m. on, and such excitement was created as only can start—a position made to approximate conditions prevailing in commercial operation. Pils, Pons, Lafon, Rouquet, Vanhoren and Bremonet took off in the order indicated, but Pils, on the last lap, Bremonet, had difficulties with a rudder spring before he had gone beyond the aerodrome, and so he retired and landed.

The other four pilots had no trouble whatever, all landing without any accident at their aerodrome, near Lezay, Bremonet as the winner, arriving first, followed by Vanhoren and Rouquet. The following day the return flight was likewise completed by all four pilots, who again finished in the order indicated for the first leg of the race. Vanhoren, on the Caudron C-25, however, had trouble with his fuel pump and had to land on route, which eliminated him from the flying.

Following are some interesting technical data on the three men who placed:

First—Bessie/4-4 on Ponce F30 (750 hp. Raynaud, Lauchla radiator). Fuel consumption, 53.77 kg. per hour, 49.4 kg. Fuel consumption per hp. per hour, 47.5 gram/kg. Second—Rouquet on Ponce VIII (90 hp. Anzani). Fuel consumption, 94.35 kg. Pay load, 154 kg. Fuel consumption per hp. per hour, 41.6 gram/kg.

Third—Dishier on Caudron C-25 (100 hp. Hispano). Fuel consumption, 154.43 kg. Pay load, 30 kg. Fuel consumption per hp. per hour, 32.5 gram/kg.

Thus, clearly following in victory in the "Grand Prix de la Haute-Aviation," the Ponce-Solair combination, even when was victorious. Bessie/4-4 on the Ponce F30 also was first place in the new European competition known as the "Grand Prix de Paris," held last December, this ship is now ranked as the first of the most practical and economical light-weighted airplane for mail and passenger transport.

Bremonet, the winning pilot, received in addition to the Cup a cash prize of \$ 36,000, Rouquet a cash prize of \$ 6,000, and Dishier a cash prize of \$ 3,000. They each drove, which were not awarded, will be used to increase next year's awards.

## Helicopter Calculations

Much serious thought and money is being spent these days on the helicopter or direct lift machine. Sir George H. D. Brink, our member, stated recently in the House of Commons that to date the British government has spent £17,000 on the development of the helicopter proposed by Louis Bréguet, an idea, in view of the size at which the experiment had arrived it would be a pity to discontinue them. The money spent on the Bréguet helicopter is additional to the £200,000 prize which the British government offer for a helicopter capable to be held in the system of 1925, and from which the Bréguet machine is excluded.

In France the government has also spent much money on calculating the helicopter, the French and the French, the Peugeot, the latter of which it recently purchased after extensive trials. Finally, in this country about \$200,000 was spent by the Army Air Service on the development of the helicopter for the helicopter.

In spite of the evident interest which attaches to this type of hovercraft or air craft, simple formulas are not readily available for calculating the probable performance of a helicopter. This is largely due to the fact that helicopters are still in the development stage, and calculations are necessarily empirical.

M. H. Oshkosh, the inventor of the helicopter which has been his basis, in a recent communication to L'Aviation of Paris furnished some highly interesting particulars on the recent study of a helicopter. To determine helicopter per-

formance, he adopts the following formula, derived from the one proposed by Colonel Renault: "For a given propeller the weight lifted per horsepower is inversely proportional to the square root of the load."

For instance, if a propeller lifts 100 kilograms with 16 hp, giving a "propeller loading" of 6.25 kilograms per hp, and it is desired to have this propeller lift 200 kilograms then the propeller loading  $P$  will be governed by the following formula:

$$P = \frac{100}{\sqrt{2}} = 7.5 \text{ kg./hp}$$

(from which the load horsepower required to lift 200 kilograms is expressed by the formula)

$$HP = \frac{200}{7.5} = 26.4 \text{ hp}$$

For a load of 200 kilograms the formula will read

$$P = \frac{200}{\sqrt{2}} = 6.75 \text{ kg./hp}$$

which gives a total horsepower required of 52.8 for a load of 200 kilograms.

On the other hand, if the load lifted with this same propeller is only 50 kilograms, the power loading equals  $16 \times \sqrt{2} = 22.1 \text{ kg./hp}$ , and the horsepower required will be 35 hp.

From these formulas M. Oshkosh has compiled the following table, which he experimentally found to be accurate within 2 per cent:

Load in kilograms	Power loading in kg./hp.
100	7.50
150	6.25
200	5.00

For two geometrically similar propellers of different diameters, the load carried with the same power loading varies to the square of the propeller diameter. For instance, if a propeller of 1 meter diameter lifts 25 kilograms with 6.25 hp, or 16 kg./hp, then a geometrically similar propeller of 2 meters diameter will lift  $25 \times 4 = 100 \text{ kg.}$  with the same power loading of 16 kg./hp.

On the basis of this formula the following table shows, for the same power loading, the total load carried with different propeller diameters:

Propeller diameter in meters	Power loading in kg./hp.	Maximum load in kilograms
1	16	25
2	16	100
3	16	225
4	16	400
5	16	625

It follows from these calculations that it is advantageous to fit a helicopter with large propellers turning slowly, or to use the largest number of propellers that may be fitted to a machine, since this will also retain a relative light loading and therefore an advantageous loading for the propeller. It also follows, however, that the propeller diameter for helicopter has a practical maximum limit of 5 or 7 meters. It should be noted, however, that M. Oshkosh is of the opinion that as the future propellers of smaller diameter at higher loading per horsepower will be possible without a corresponding loss in efficiency. Reference in Table II shows that with a single propeller of 4 meters diameter lifts 80 kilograms with 6.25 hp, four propellers of 2 meters diameter only lift 80 kg. at a greater weight of 6.25 kilograms, although the power expenditure in the latter case is 42.2 hp.

## Light Plane Flies High

A Ponce light plane piloted by Louis Couper, the winner at 1929, was once offered by a Paris daily for a light plane race, on Aug. 8 run to an altitude of 2100 meters in 1/2 to 30 sec. The performance was officially reported by the Aero Club of France.

# Principles of Efficient Aileron Design\*

## Fundamental Requirements for Effective Lateral Control

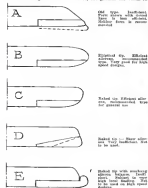
By W. S. DECHER

Recent data have shown that certain types or types of ailerons in extreme use, are in reality quite inefficient and simply wasted for the high speeds now attained.

The most important of the characteristics required of ailerons are: (1) Effectiveness under all conditions of flight; (2) Small moments about the hinge; (3) High efficiency (avoid giving moment opposing turn); (4) Simplicity in construction.

## Chord, Span and Area

Tests conducted at the National Physical Laboratory (British A.C.A. 8, 4 N. 550 and 635) show that the maximum lift moment obtainable is practically independent of the aileron chord, provided that it is not less than about 15 per cent of the wing chord  $c$ . However, the greater the value of



Various types of aileron used on airplanes

$d/c$ , the greater the moments about the hinge, the greater the rolling moments (opposing turn), and the less the rolling is obtained under all  $d/c$  is between 0.20 and 0.50.

The ratio of rolling moment to hinge moment, usually called aileron efficiency, is found to be a maximum when  $d/c = 0.75$ . For the best average results  $d/c$  should be greater than 0.25 and less than 0.75. Very long ailerons are liable to deflect and bend at the hinges.

\*Excerpt from R.A.C.A. Technical Note No. 249. Originally prepared by Technical Note No. 249, Bureau of Aeronautics, Navy Department.

For the average airplane the aileron area is about 15 per cent of the wing area. This would correspond to an aileron with  $d/c = 1/3$  and  $L/S = 1/3$ .

These proportions were used in ailerons used, although the present literature is toward narrower ailerons.

Too much aileron area is used in many cases, particularly with the older designs in which  $d/c$  is large. There is a well defined lower limit to the amount of aileron area required for a given degree of lateral control, in this case, corresponding to  $d/c = 0.25$  and  $L/S = 0.50$ .

## Plan Form

The plan form of the ailerons is perfectly fitted by the plan form of the wing. It is shown that the best results are obtained from a wing tip provided slightly, or related with the leading edge longer than the trailing edge. The wing tip should never be made with the leading edge shorter than the trailing edge for several reasons. The most important of which is the extremely high loading which occurs on the extreme tip of this type of wing. With the ordinary construction this peak in loading occurs on the aileron and represents the hinge moments which determine the aileron efficiency. Another reason for avoiding the wing tip related to the leading edge is shorter than the trailing edge, may be found in the behavior of the general pressure distribution on the wing tip. It is well known that a slight vacuum in angle of attack toward the tip imposes the pressure of the wing by preventing the entire breakdown in lift which will follow close on the tip, and thus reducing the loading over the wing. It may easily be seen that the old type of aileron shown as A in Fig. 1, gives an increase in moment toward the tip and that the new one may be seriously decreased.

The best plan form for an aileron is not so far definitely determined, although it is known that certain forms such as B and C are better than others. These forms are recommended for general use.

The "saw" pattern is objectionable when the lift is shown in it. The full effect of the negative moment is lost and the loading is objectionable. It is usually a failure to be met with this form of aileron loading when the wing defects. The hinges of all ailerons should be so arranged as to maximize the effect of any warp or twist in the wing.

## General Conclusions

The following conclusions may be drawn from a study of the references given below:

- (1) The aileron chord should be about 20 per cent of the wing chord—never more than 33 per cent and never less than 10 per cent. It is recommended that 30 per cent be used as the proper limit.
- (2) The aileron span should be greater than 20 per cent of the semi-wing span. With an aileron having  $l/s = 0.25$ , the span  $l/s$  is between 48 per cent and 96 per cent of the semi-wing span for normal control when the wing has an aspect ratio of 6.
- (3) In the general case the aileron area should vary from 10 per cent to 15 per cent of the wing area or the aileron chord values from 20 per cent to 30 per cent of the wing chord.
- (4) The plan form of the aileron should be such that there is an effect a vacuum of nearly at least toward the tip—such as that given in a circular aileron or an elliptically rounded wing tip.
- (5) The aileron should never extend beyond the mean tip of the wing.
- (6) All types of aileron settings are to be discouraged.
- (7) The aileron hinges should be designed to prevent bending of the wing defects on turns.
- (8) Ailerons on high speed airplanes should always be of

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